Rascal Requirements Verification Document

Saint Louis University

Rascal



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Copper Operational

Test Plan

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# INTRODUCTION

The Rascal Requirements Verification Document serves to define and elaborate on each of the requirements laid out in the RCL-RVM-CMQA1 Rascal Requirements Verification Matrix (RVM), as well as the ways in which each of said requirements will be validated prior to and during the Rascal mission.

The Rascal Mission, as defined by the RVM, can be broken down into Six Stages, with each stage having a set of requirements directly associated with it. Within the requirements associated with each of these stages are sub-requirements associated with a particular subsystem of the Rascal CubeSat system. The manner in which each of these requirements is associated with a particular stage or subsystem is with a simple notation scheme, as shown in Table 1.

**Table 1. RVM Requirement Notation**

|  |  |  |  |
| --- | --- | --- | --- |
| **Mission Identifier** | **Stage Identifier** | **Sub-System Identifier** | **Requirement Number** |
| RCL | Pre Launch (PL) | Structures (STR) | Number Order |
| Post Launch Ejection (PLE) | Thermal (THM) |
| Separation and Stabilization (SS) | Propulsion (PRP) |
| Stationkeeping (SK) | Testing (TST) |
| “Escape” (ESC) | Mission Operations (MOP) |
| Rendezvous (RDZ) | Payload (PLD) |
| **Example** | **RCL.PL.STR1 (1st Pre-Launch Structures Requirement)** | | |

Ultimately, the goal of this document is to provide a specific rubric from which to develop the Rascal mission such that it meets all of the design constraints and mission success criteria laid out in the RCL-P-CMQA2 Rascal Request for Proposal document. Thus, if the designed mission meets all of the requirements laid out in this document, it can be considered to have successfully executed the Rascal mission as a whole.

# NOMENCLATURE

*RVM* Requirements Verification Matrix

*RFP* Request for Proposal

*SSRL* Space Systems Research Lab

*PL* Pre-Launch

*PLE* Post-Launch Ejection

*SS* Separation and Stabilization

*SK* Stationkeeping

*ESC* “Escape”

*RDZ* Rendezvous

*STR* Structures Subsystem

*THM* Thermal Subsystem

*PLD* Payload Subsystem

*PRP* Propulsion Subsystem

*TST* Testing Subsystem

*MOP* Mission Operations Subsystem

*CMQA* Configuration, Management, and Quality Assurance Subsystem

*U* Standard Unit (10 cm x 10 cm x 10 cm)

*Jade* Sub-Satellite of Rascal System #1

*Ruby* Sub-Satellite of Rascal System #2

*GEVS* General Environmental Verification Specification

*NASA* National Aeronautics and Space Administration

*RF* Radio Frequency Communication

*CDS* CubeSat Design Specification

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# REQUIREMENT STAGES

# Pre-Launch Requirements

1. **The Total CubeSat System Volume Shall Not Exceed 6U**

This requirement comes from the Team Bravo RFP (RCL-P-CMQA2) Section 2.1 Mission Objective, as well as from the CubeSat Design Specification (CDS) Document, Rev 12, Section 2.2 Mechanical Requirements. This requirement will be met by verifying the maximum outer dimensions of the CubeSat system fall within those dictated by a 6U architecture (20 cm x 10 cm x 300 cm), as well as performing a fit check with a 6U deployer, thus falling under the Examine validation method.

1. **The Total CubeSat System Mass Shall Not Exceed 8.0 kg**

This requirement stems from the CDS Document, Rev 12, section 2.2.16 mass definition for 3U CubeSats. This requirement will be satisfied by weighing the completed spacecraft prior to integration with its deployer and ensuring that its mass is less than 8.0 kg. The requirement validation method falls under the Examine category.

1. **All Materials Used in the CubeSat System shall have a Total Mass Loss of Less Than 1.0%**

This requirement derives from the CDS Document, Rev 12, section 2.1.7.1. The spacecraft must satisfy all low-outgassing criteria as to prevent contamination of other spacecraft and the launch vehicle during testing, integration, and launch. This requirement will be satisfied by using only materials found on the NASA approved list of low outgassing materials, as specified on <http://outgassing.nasa.gov>. Thus, the validation method associated with this requirement falls under the Analyze category.

1. **All Materials Used in the CubeSat System Shall Have a Collected Volatile Condensable Material of less than 0.1%**

This requirement is dictated by the CDS Document, Rev 12, section 2.1.7.2. The spacecraft must satisfy all low-outgassing criteria to prevent contamination of other spacecraft and the launch vehicle during testing, integration, and launch. This requirement will be satisfied by using only materials found on the NASA approved list at <http://outgassing.nasa.gov>. Thus, the validation method associated with this requirement falls under the Analyze category.

1. **The CubeSat System Must be capable of Operating in Orbit for at Least 6 Months**

This requirement comes from the Team Bravo RFP (RCL-P-CMQA2) Table 1-2, Proposed Mission Constraints. This requirement will be satisfied by constructing an accurate power budget, performing a battery cycle test to ensure that the batteries used to power the CubeSat system can charge and discharge correctly, performing a day-in-the-life test of the integrated CubeSat system as to verify the accuracy of the power budget developed for the mision, performing a solar panel charge test as to verify that the solar panels are performing as designed and can charge the CubeSat power system, and performing a solar cell degradation analysis to determine how quickly the solar cells will degrade due to radiation damage and determine how much power margin remains after six months of operation. Along with this, an orbit must be selected that allows for the Rascal CubeSat system to remain in orbit for the duration of this time period. Thus, the validation method for this requirement falls under the Analysis category.

1. **The CubeSat System Must Deorbit within 25 Years of being Launched**

This requirement stems from the Process for Limiting Orbital Debris Document (GSFC-STD-7000A), which requires that all objects put in orbit around the Earth deorbit within 25 years, as to stymie the accumulation of orbital debris that permanently resides in LEO and prove a continuous risk to satellite development and survival. This requirement will be satisfied by performing an orbital analysis using orbital parameters provided by the launch provider to calculate the orbital lifetime of the CubeSat system, as to verify that it will deorbit within the allotted time limit. If it is determined that the designed CubeSat system cannot meet said requirement, either a new launch vehicle will need to be selected or a deorbit mechanism will need to be incorporated into the CubeSat system’s design. Thus, the validation of this requirement falls under the Analysis category.

1. **Jade and Ruby Shall be Conjoined Prior to Launch Vehicle Integration**

This requirement comes from RCL.PL.STR1, which constrains the CubeSat system to a 6U volume.. Because the target vehicle must be incorporated into the 6U volume, it is necessary that the two spacecraft be securely conjoined prior to launch vehicle integration. This requirement will be satisfied by conducting an integrated vibration test, as discussed in requirement RCL.PL.TST1, and a release mechanism separation test, as discussed in requirement RCL.PL.STR16. Thus, the validation of this requirement falls under the Test category.

1. **The CubeSat System Shall Incorporate a Remove Before Flight Pin**

This requirement comes from the CDS Document, Rev 12, section 2.3.4, which dictates that a Remove Before Flight (RBF) pin cuts off all power when inserted into the CubeSat system by physically separating the CubeSat power supply from the rest of the CubeSat system, as well as be accessible from the deployer’s access points, which are shown in Figure 1-1. This is done so that the spacecraft is not active during testing and deployer integration. Thus, the validation of this Requirement falls under the Demo requirement.



**Figure 1-1. Location of Remove Before Flight Pin Access Points**

1. **The CubeSat System Shall Incorporate a Deployment Switch**
2. **No Protrusion Shall Extend beyond 6.5 mm Normal to Any External Surface of Jade or Ruby**
3. **No External Components Other than the CubeSat Rails of Jade and Ruby may make Contact with the Deployer**
4. **The Deployer Shall not be Used to Secure Any CubeSat Deployables**
5. **The Center of Gravity of the total CubeSat System Shall be Located within a Sphere of 2 cm of the Geometric Center of the System**
6. **The Center of Gravity of Jade and Ruby Shall be Located within a Sphere of 2 cm of their Geometric Center**
7. **The CubeSat System Coordinate System Shall be Defined As Specified in Figure 1-1**
8. **The Local Coordinate System of Jade and Ruby Shall be Defined as Specified in Figure 1-1**
9. **The Ends of the Rails on the +Z/-Z Faces of the CubeSat System Shall have a Minimum Surface Area of 6.5 mm x 6.5 mm**
10. **The +Y/-Y Faces of Ruby and Jade Shall have a Length of 100 mm**
11. **Jade and Ruby Shall be Capable of Determining Relative Displacement between Each Other**
12. **The CubeSat System Shall be Capable of Recording Relative Displacement Data between Jade and Ruby**
13. **Low Friction, 2D Testing of the CubeSat System Release Mechanism Shall be Conducted**
14. **All Pressure Vessels Shall have a Factor of Safety of No Less Than 4**
15. **All CubeSat Components Shall be Rated to Operate within Temperature Range of at least -20⁰C to 70 ⁰C**

This requirement is set by Team Bravo's RFP, Section 1. The requirement was created to ensure that all components will survive the temperature range that the CubeSat system will encounter on orbit. To verify that this requirement has been met, an Analysis of the data sheets for each component shall be performed to check that the survival temperatures fall in this range. For those components that are developed at the SSRL, each component that is used in its assembly will be rated to operate within said range. Thus, the validation of this requirement falls under the Analysis category.

1. **Static Thrust Testing Shall be Performed with the Flight Version of All Pressure Vessels at a Pressure No Greater than 1x10-4 Torr Prior to CubeSat Integration**

This requirement is set by Environmental Testing Requirements. This requirement was created to ensure that the propulsion system will operate in vacuum environment. To verify that this requirement has been met, the propulsion system must pass a static thrust test with no anomalies in a vacuum chamber at a pressure no greater than 1x10-4 Torr. This data will be critical in determining the performance of a given propulsion design, and thus whether or not it will be able to meet all of the requirements laid out in this Requirements Verification document. If a design does not meet these requirements, it will be necessary to either change the design or re-evaluate the actual requirements. Thus, this requirement will be validated through Testing.

1. **All Pressure Vessels Must Pass Thermal Cycle Testing between Temperatures of -30 ⁰C and 70 ⁰C for at Least Two Cycles or for 10 Hours**

This requirement is set by Environmental Testing Requirements. The requirement was created to ensure the propulsion system will survive the temperatures that it will encounter while it is in orbit. The temperature profile that will be used was established by previous Educational Launch of Nanosatellites (ELaNa) missions to help validate the performance of CubeSat missions in space-like thermal environments. An illustration of this particular profile is shown in Figure 1-2 below. To verify that this requirement has been met, the propulsion system shall perform static thrusts before the thermal cycle, to establish that the system works before the test. Then a static thrust test shall occur at various points throughout the thermal cycle, with data being collected on the performance of the system for each thrust. Finally, a static test shall take place after the thermal cycle, as to check that the propulsion system still operates under normal conditions. Thus, the validation of this requirement falls under the Testing category.



**Figure 1-2. Rascal Thermal Cycle Test Profile**

1. **Low Friction, 2-D Dynamic Thrust Testing Shall be Conducted with All Pressure Vessels**
2. **The CubeSat System Must Survive Random Vibration Testing Relative to the NASA GEVS Qualification Profile**

This requirement is set by the NASA document General Environmental Verification Standard for GSFC Flight Programs and Projects. The requirement was created to ensure the CubeSat system will survive the vibration environment of launch. The GEVS profile was created as a general vibration profile to cover as many vibration environments as possible, and is shown in Figure 1-3.

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**Figure 1-3. NASA GEVS Random Vibration Profile**

To verify that this requirement has been met, the CubeSat System shall perform an abbreviated functional test to establish that all systems work before testing. A sine sweep shall be performed on before each axis of testing, as to establish the CubeSat System's natural response to vibration, thus allowing for failure assessment before the beginning of each axis of testing. Then the CubeSat system shall be shaken at the GEVS random vibration qualification profile for each axis. After the CubeSat system has been shaken on all three axes, a full functional test will be performed to check that all systems survived the process and are performing as normal. Thus, the validation method for this requirement falls under the Testing category.

1. **The CubeSat System Shall be Subjected to a Temperature of 60 ⁰C at a Pressure No Greater than 1x10-4 Torr for a Minimum of 6 Hours**

This requirement exists to ensure the CubeSat system will accepted for launch vehicle integration, with its origin stemming from previous ELaNa missions that have required its execution before said integration*.* The requirement was created to ensure the CubeSat system does not release volatiles on orbit, which could damage nearby spacecraft. To verify this requirement has been met, an abbreviated functional test will be performed before bakeout to check that all the systems work. Then the CubeSat system shall undergo the bakeout at a temperature of 60 °C and a pressure no greater than 1x10-4 torr for at least six hours. After the bakeout other abbreviated functional test shall be performed to check that all systems survived bakeout. This process ensures that if any volatile material were to be emitted in orbit that it instead is emitted during bakeout testing. Thus, the validation of this requirement falls under the Testing category.

1. **The CubeSat System Shall be Able to Execute All Commands Associated with its Operation over RF**
2. **The CubeSat System Shall be Able to Close a Link with the SSRL Ground Station from a Distance of at least 200 meters**
3. **The CubeSat System Shall be able to Document the Functionality of Each of its Subsystems through the Running of a Full-Functional Test**

# Post-Launch Ejection Requirements

1. **The CubeSat System Shall not Broadcast in RF Until Ejection +45 Minutes**
2. **The CubeSat System Shall not Release Deployables Until Ejection +45 Minutes**
3. **The CubeSat System Shall Establish Communication Between Itself and the SSRL Ground Station**
4. **The CubeSat System Shall Pass a Health Check Administered from the SSRL Ground Station**

# Separation and Stabilization Requirements

1. **Jade and Ruby Shall be Capable of Separating from One Another with a Relative Velocity of No Greater than 5 cm/s**
2. **Jade and Ruby Shall Achieve a Local Slew Rate of Less than 1 deg/s**
3. **Jade and Ruby Shall Continuously Record Relative Displacement Data Between Each Other**

# Stationkeeping Requirements

1. **Jade and Ruby Shall be Able to Stationkeep within a 10-75 meter Sphere of Each Other for at Least 5 Orbits**

# “Escape” Requirements

1. **Jade and Ruby Shall be Able to Perform an “Escape” Maneuver that Increases the Relative Displacement Between Each Other to at Least 100 Meters within 1 Orbit**

# Rendezvous Requirements

1. **Jade and Ruby Shall be Able to Perform a Rendezvous by Decreasing the Relative Displacement Between Each Other to Within 50 meters for at Least 5 Orbits**